

Approach for Uncertainty Propagation and Robust Design in CFD Using Sensitivity Derivatives

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Motivation

- **Uncertainty Analysis**
 - Typically neglected in CFD analysis
 - May play an important role in system design
 - Input parameter uncertainty propagation via approximate statistical moments feasible
 - Requires efficient calculation of sensitivity derivatives
(Taylor et al., 2001)
- **Robust Optimization**
 - Deterministic optimization for discrete operating conditions may produce inadequate results
 - Robust gradient-based optimization feasible using approximate statistical moments

Outline

- CFD problem with input parameter uncertainty
- Uncertainty propagation:
 - Approximate statistical moment method
(first moment = mean, second moment = variance)
 - Applications of uncertainty analysis to quasi 1-D Euler
- Robust design
 - Robust design using approximate statistical moments
 - Applications of robust design to quasi 1-D Euler
- Conclusions

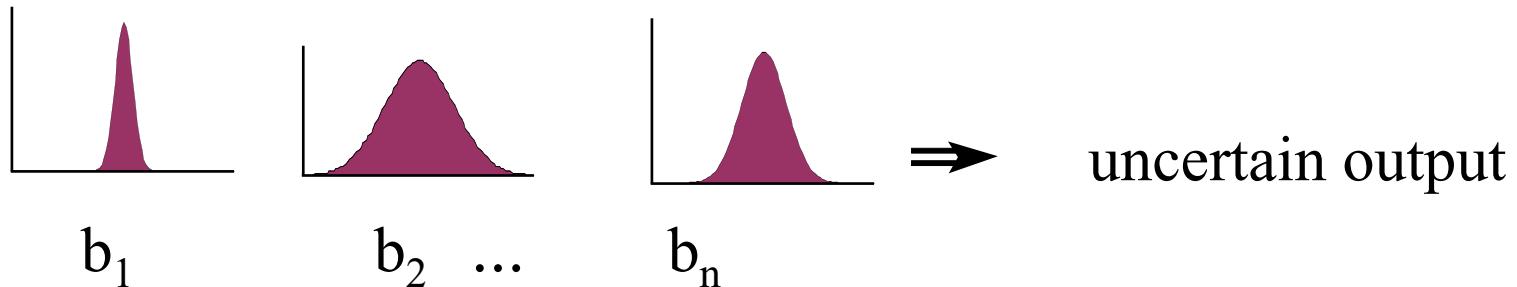
CFD Problem with Input Parameter Uncertainty

Deterministic Problem: Parameters precisely known

Non-Deterministic Problem Parameters not precisely known

Input parameters assumed:

- statistically independent
- random
- normally distributed about mean
- mean values, $\bar{\mathbf{b}} = \{\bar{b}_1, \dots, \bar{b}_n\}$
- standard deviations, $\sigma_{\mathbf{b}} = \{\sigma_{b_1}, \dots, \sigma_{b_n}\}$



Uncertainty Propagation Using Approximate Statistical Moments

1st-and 2nd- Order Taylor Series Approximations for Output F(**b**)

First-Order: $F(\mathbf{b}) = F(\bar{\mathbf{b}}) + \sum_{i=1}^n \frac{\partial F}{\partial b_i} (b_i - \bar{b}_i)$
(FO)

Second-Order: $F(\mathbf{b}) = F(\bar{\mathbf{b}}) + \sum_{i=1}^n \frac{\partial F}{\partial b_i} (b_i - \bar{b}_i) +$
(SO)
 $+ \frac{1}{2!} \sum_{j=1}^n \sum_{i=1}^n \frac{\partial^2 F}{\partial b_i \partial b_j} (b_i - \bar{b}_i)(b_j - \bar{b}_j)$

where all derivatives are evaluated at the mean values, $\bar{\mathbf{b}}$.

Approximate Mean and Variance

FO FM: $\bar{F} = F(\bar{b})$

FO SM: $\sigma_F^2 = \sum_{i=1}^n \left(\frac{\partial F}{\partial b_i} \sigma_{b_i} \right)^2$

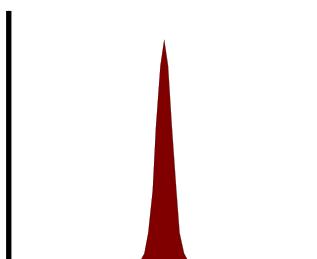
SO FM: $\bar{F} = F(\bar{b}) + \frac{1}{2!} \sum_{i=1}^n \frac{\partial^2 F}{\partial b_i^2} \sigma_{b_i}^2$

SO SM: $\sigma_F^2 = \sum_{i=1}^n \left(\frac{\partial F}{\partial b_i} \sigma_{b_i} \right)^2 + \frac{1}{2!} \sum_{j=1}^n \sum_{i=1}^n \left(\frac{\partial^2 F}{\partial b_i \partial b_j} \sigma_{b_i} \sigma_{b_j} \right)^2$

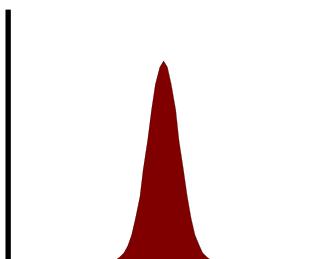
where all derivatives are evaluated at the mean values, \bar{b} .
Note second-order shift in mean.

Accuracy of Mean and Variance Approximations

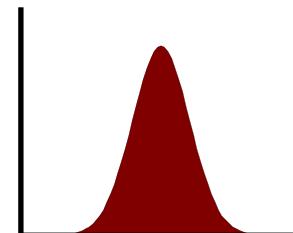
- FO and SO approximate moments compared to Monte Carlo (MC) simulation results
- Error associated with mean of Monte Carlo simulation: $\frac{\sigma_F}{\sqrt{n}}$
- Parametric study performed with increasing variation in input parameters:



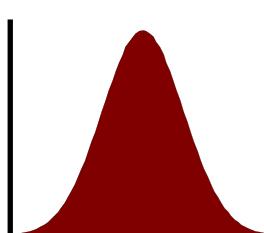
Case 1



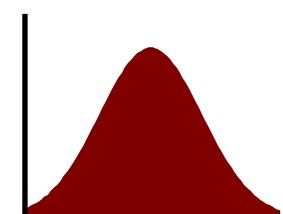
Case 2



Case 3



Case 4



Case 5

Application: Quasi 1-D Euler Problem

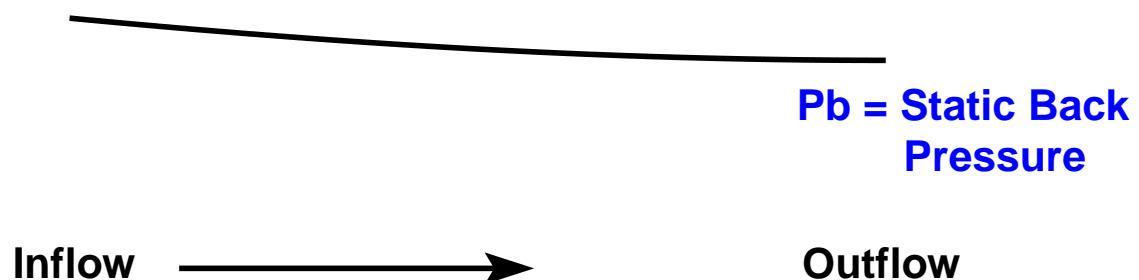
Input Random Variables:

Geometric Flow $b = \{a, b\}$
 $b = \{M_{\infty}, P_b\}$

CFD Output Function:

$F = \{M\}$

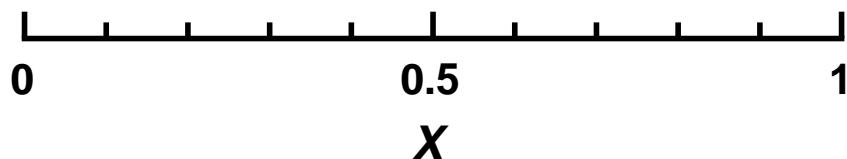
$M_{\infty} =$
Free Stream
Mach Number



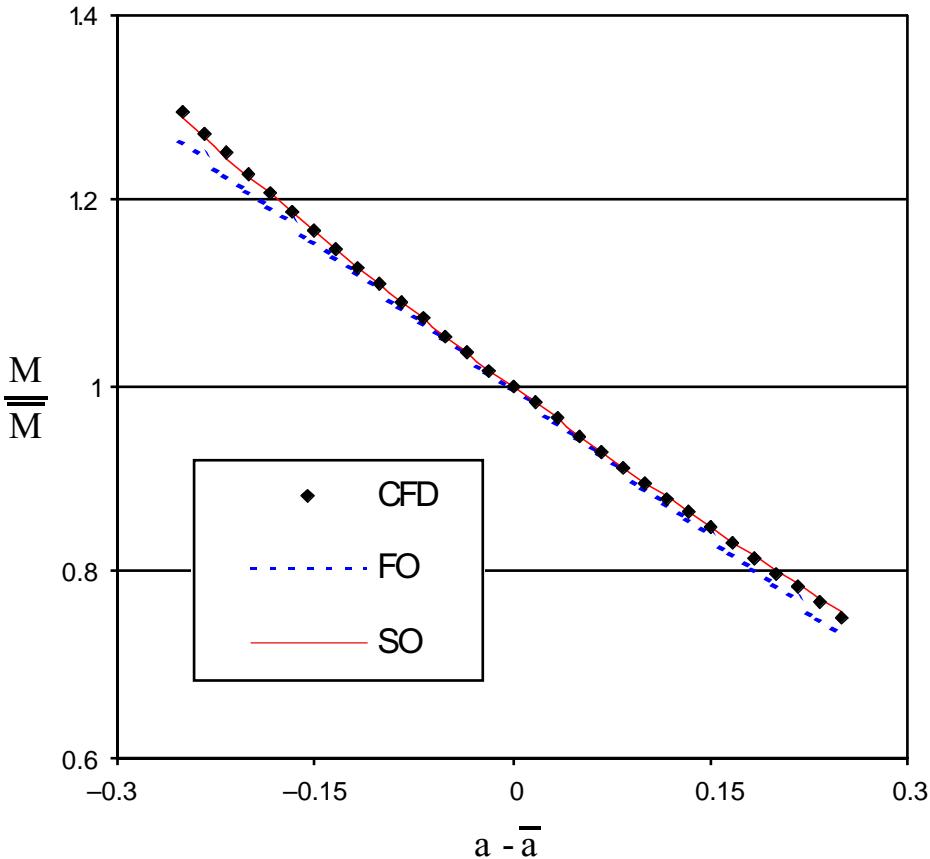
**M = Mach Number at
Nozzle Inlet**



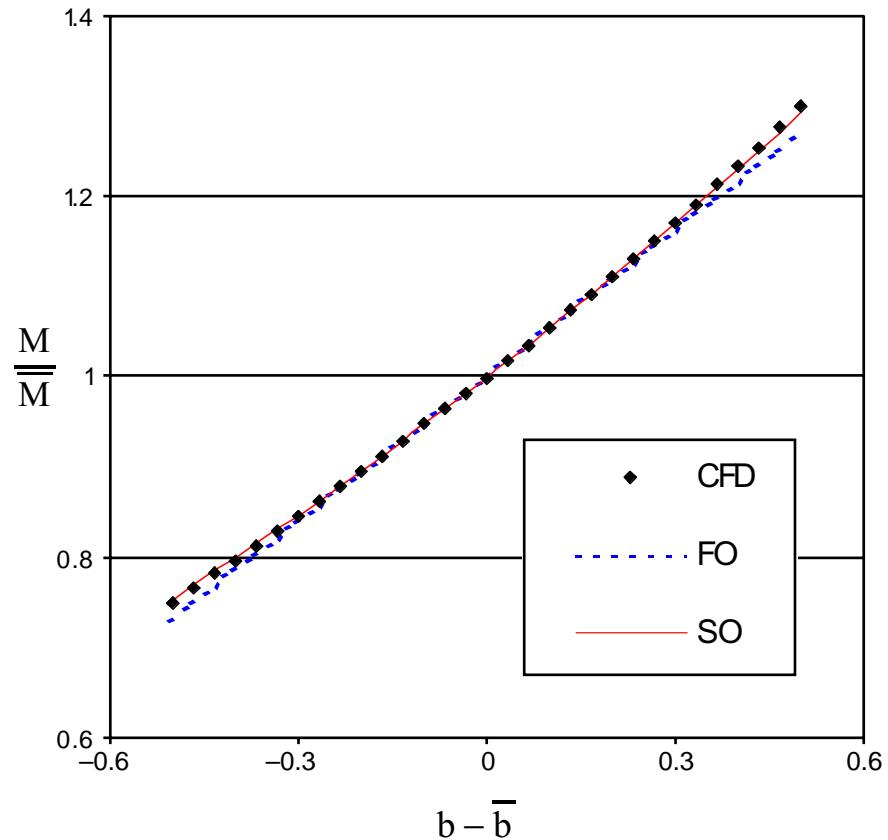
$$\text{Area} = 1 - ax + bx^2$$



Function Approximations for $F=M(a,b)$

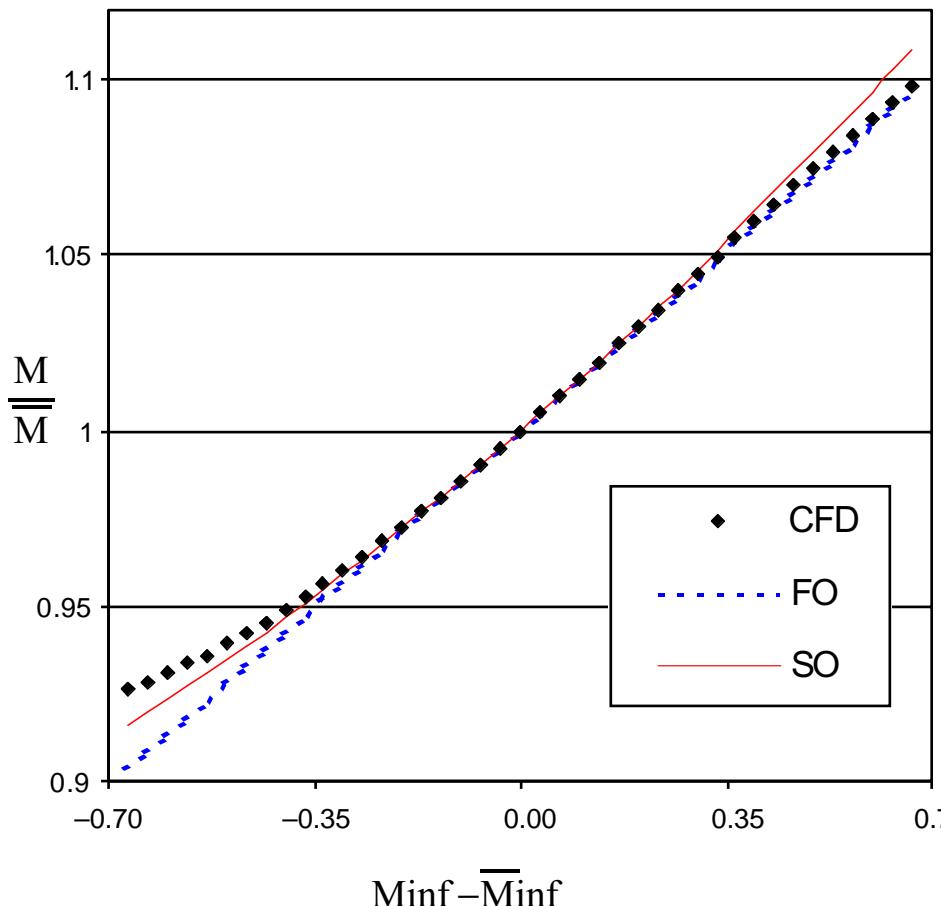


Comparison of Function Approximations
vs. CFD Solution, $b = \bar{b}$

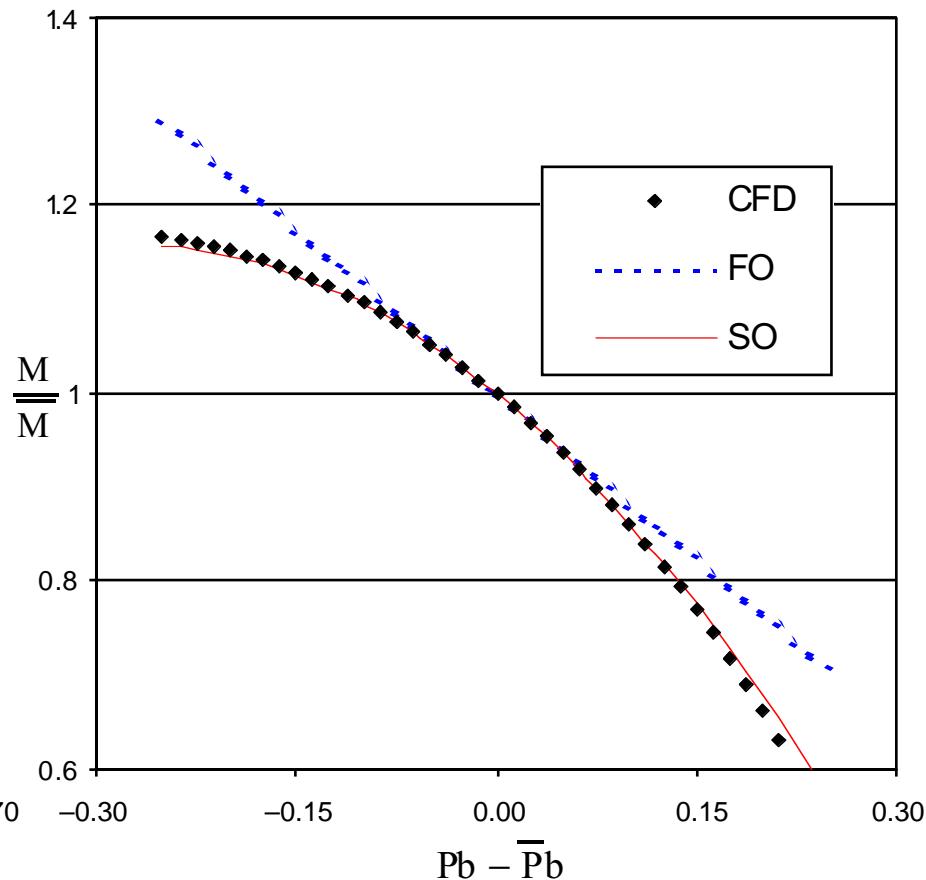


Comparison of Function Approximations
vs. CFD Solution, $a = \bar{a}$

Function Approximations for $F=M(M_{\infty}, \bar{P}_b)$



Comparison of Function Approximations
vs. CFD Solution, $P_b = \bar{P}_b$



Comparison of Function Approximations
vs. CFD Solution, $Minf = \bar{M}_{\infty}$

Example Mean and Variance Approximations

FO FM: $\bar{M} = M(\bar{a}, \bar{b})$

FO SM: $\sigma_M^2 = \left(\frac{\partial M}{\partial a} \sigma_a \right)^2 + \left(\frac{\partial M}{\partial b} \sigma_b \right)^2$

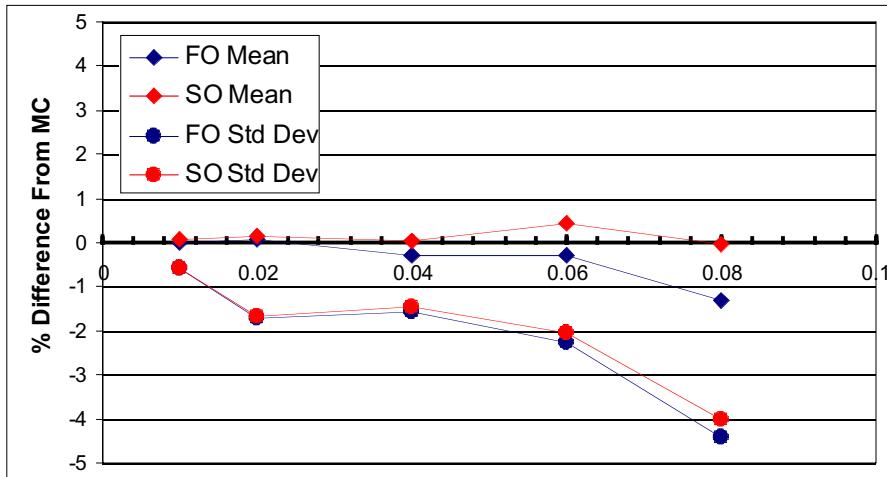
SO FM: $\bar{M} = M(\bar{a}, \bar{b}) + 0.5 \left(\frac{\partial^2 M}{\partial a^2} \right) \sigma_a^2 + 0.5 \left(\frac{\partial^2 M}{\partial b^2} \right) \sigma_b^2$

SO SM: $\sigma_M^2 = \left(\frac{\partial M}{\partial a} \sigma_a \right)^2 + \left(\frac{\partial M}{\partial b} \sigma_b \right)^2 + 0.5 \left(\frac{\partial^2 M}{\partial a^2} \sigma_a^2 \right)^2 +$

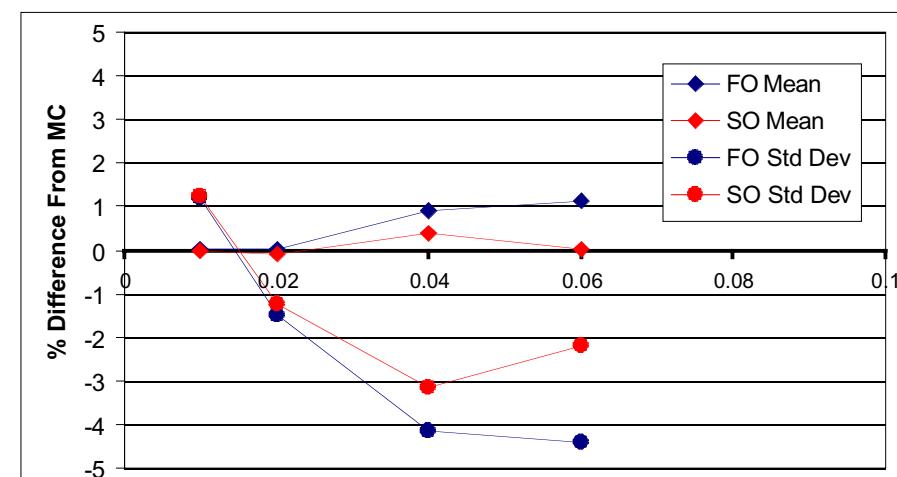
$$+ 0.5 \left(\frac{\partial^2 M}{\partial b^2} \sigma_b^2 \right)^2 + \left(\frac{\partial^2 M}{\partial a \partial b} \sigma_a \sigma_b \right)^2$$

Results: Comparison of Statistical Approximations vs. Monte Carlo Simulation

Geometric Example

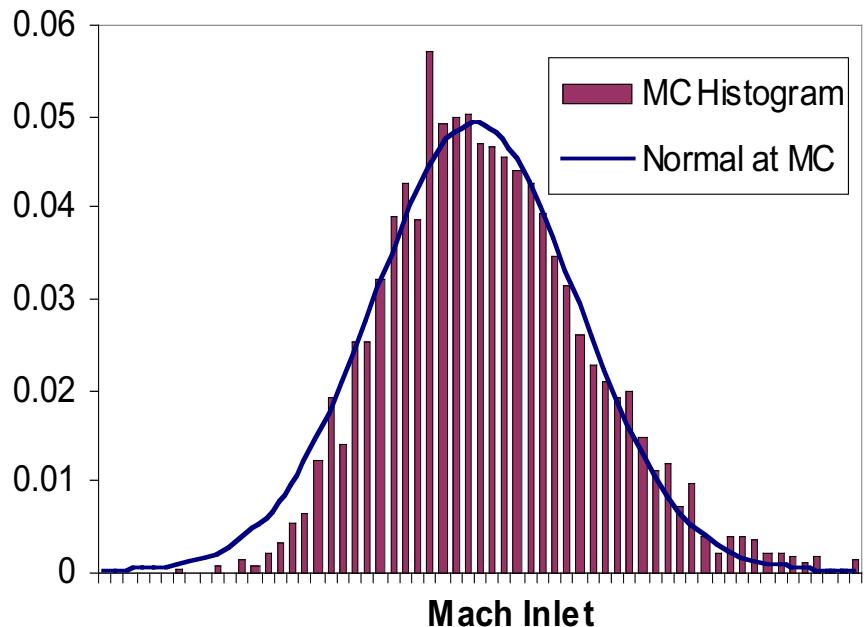


Flow Parameter Example



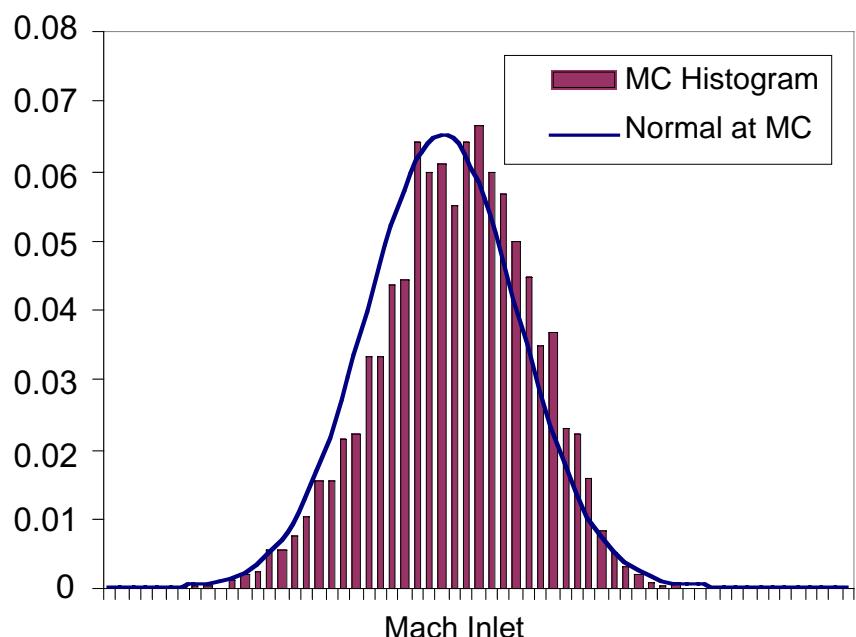
- For larger input parameter σ , second-order generally gives better predictions.
- Approximations predict first moment more accurately than second moment.

Probability Distribution Function for M(a,b) from Monte Carlo Simulation

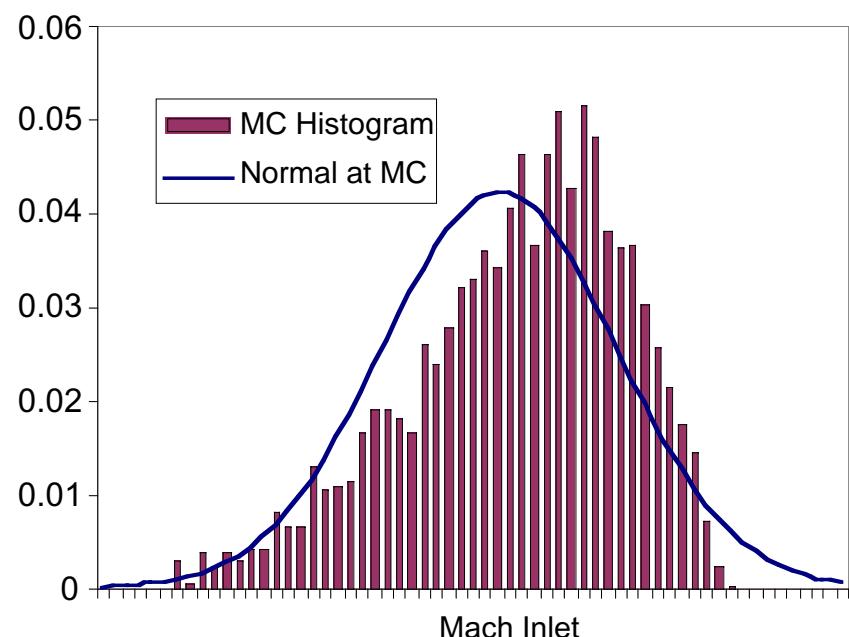


$$\sigma_a = \sigma_b = 0.08$$

Probability Distribution Functions for (M_{inf}, P_b) from Monte Carlo Simulations



$$\sigma_{M_{\infty}} = \sigma_{P_b} = 0.02$$



$$\sigma_{M_{\infty}} = \sigma_{P_b} = 0.06$$

Robust Optimization Using Approximate Statistical Moments

Conventional Optimization

Objective function is a function of the CFD output, \mathbf{F} , state variables, \mathbf{Q} , and input variables, \mathbf{b}

min Obj,
subject to

$$\text{Obj} = \text{Obj}(\mathbf{F}, \mathbf{Q}, \mathbf{b})$$

$$\begin{aligned}\mathbf{R}(\mathbf{Q}, \mathbf{b}) &= 0 \\ \mathbf{g}(\mathbf{F}, \mathbf{Q}, \mathbf{b}) &\leq 0\end{aligned}$$

where \mathbf{R} represents the flow (state) equation residuals.

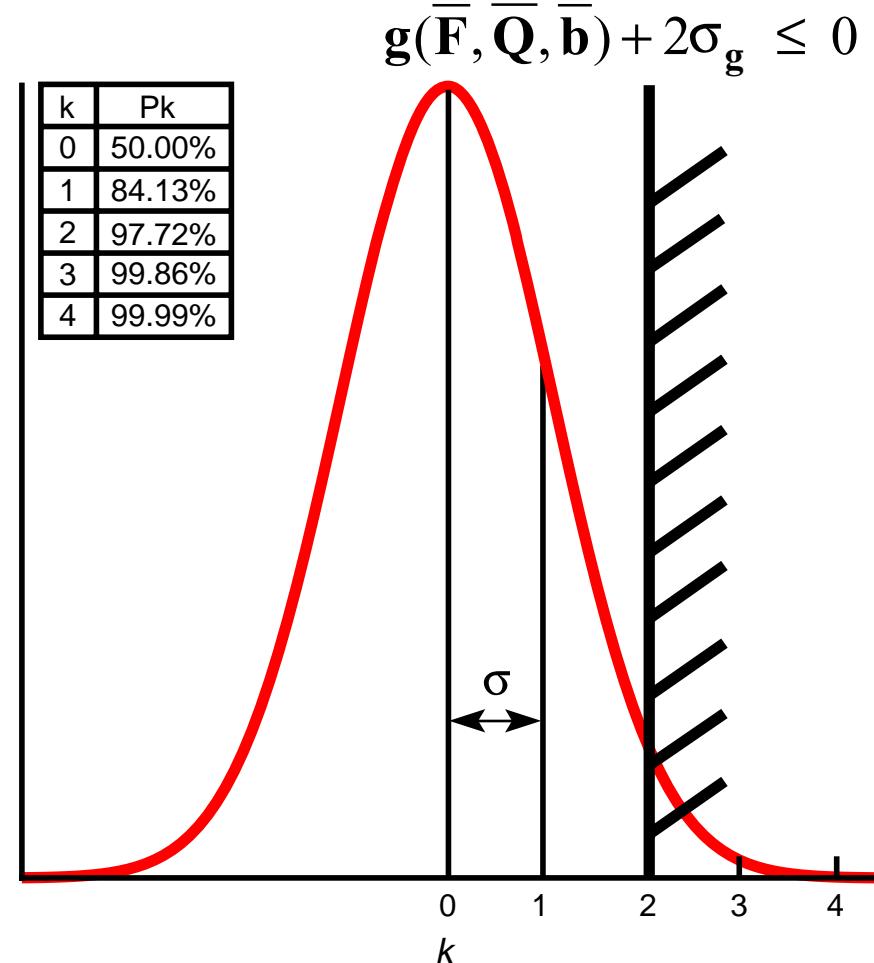
Robust Optimization

Objective function uncertain due to uncertain input variables

$\min \text{Obj}, \quad \text{Obj} = \text{Obj}(\bar{\mathbf{F}}, \sigma_{\mathbf{F}}, \bar{\mathbf{Q}}, \bar{\mathbf{b}})$
subject to $\mathbf{R}(\bar{\mathbf{Q}}, \bar{\mathbf{b}}) = 0$

$$g(\bar{\mathbf{F}}, \bar{\mathbf{Q}}, \bar{\mathbf{b}}) + k\sigma_g \leq 0$$

- $k\sigma_g$ represents the desired safety factor for probabilistic constraint satisfaction



Demonstration of Robust Optimization

Robust Shape Optimization

$$\begin{aligned} \min \text{Obj}, \quad & \text{Obj} = \text{Obj}(\bar{\mathbf{M}}, \sigma_M, \bar{a}, \bar{b}) \\ \text{subject to} \quad & \mathbf{R}(\bar{\mathbf{M}}, \bar{a}, \bar{b}) = 0 \quad \text{and} \quad V(\bar{a}, \bar{b}) + k\sigma_V \leq 0 \end{aligned}$$

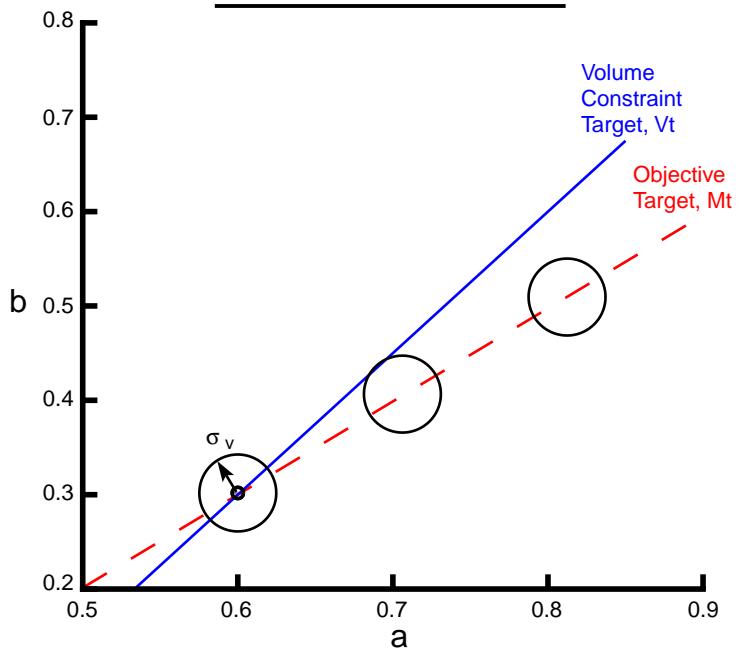
where V is a constraint involving the nozzle volume.

Target Matching Problem

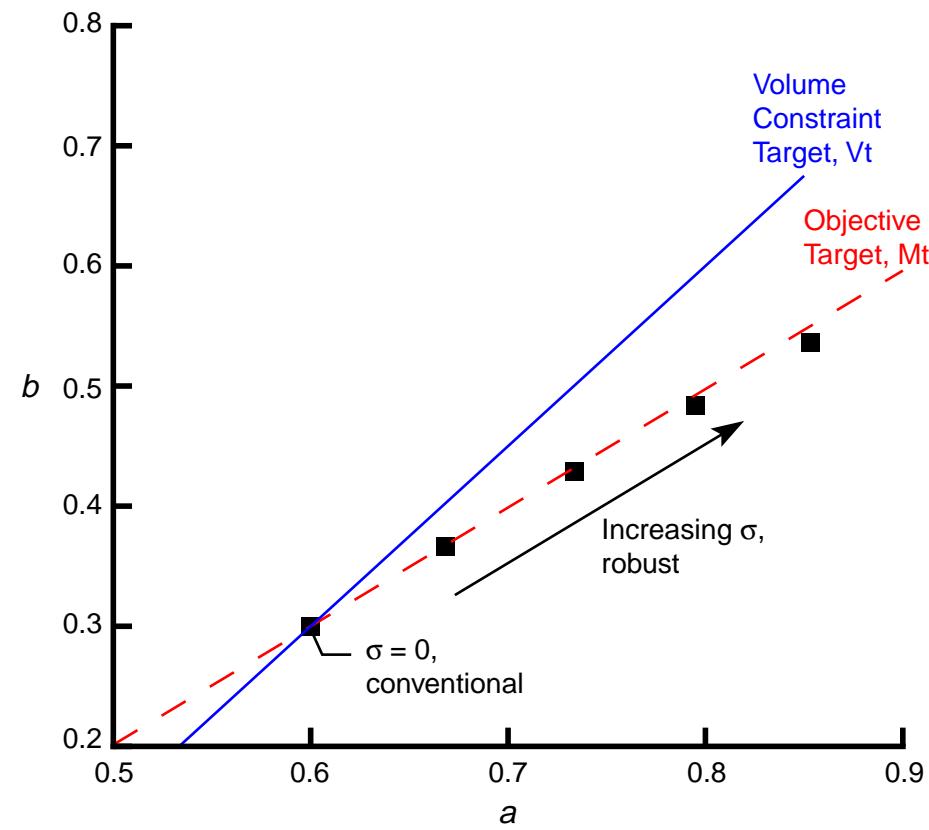
$$\text{Obj}(\bar{\mathbf{M}}, \sigma_M, \bar{a}, \bar{b}) = \{[\bar{\mathbf{M}}(\bar{a}, \bar{b}) - M_t]^2 + \sigma_M^2\}$$

$$V(\bar{a}, \bar{b}) - V_t + k\sigma_V = 0$$

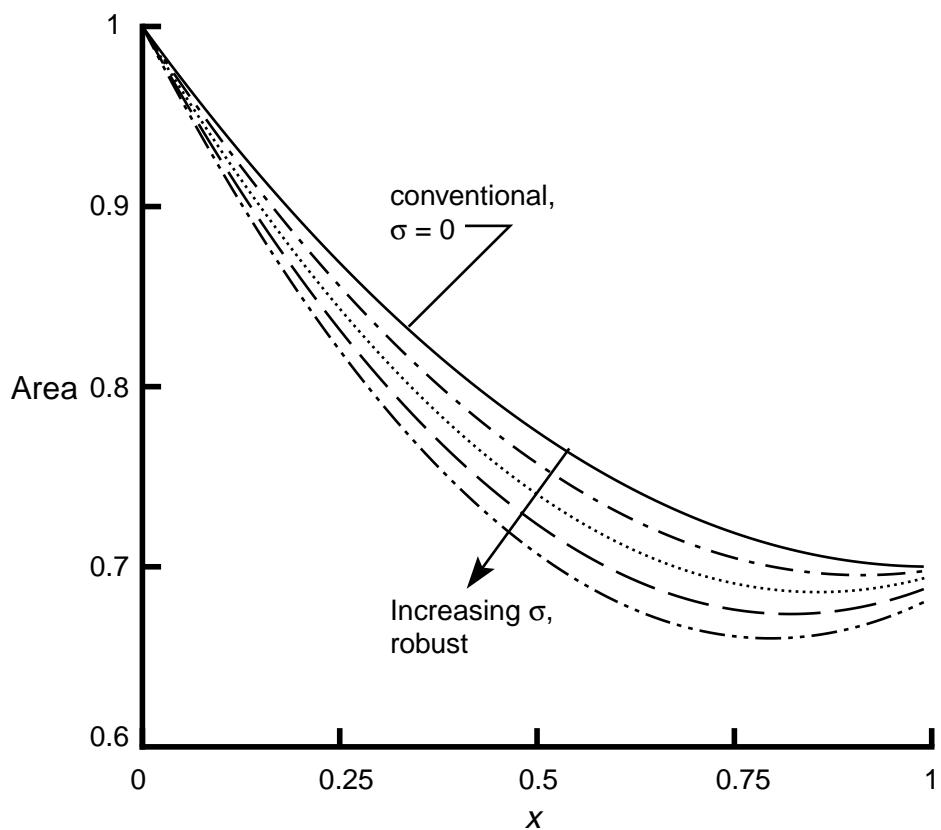
Design Space



Robust Shape Optimization Results, Increasing σ , $P_k = P_1 = 84.13\%$

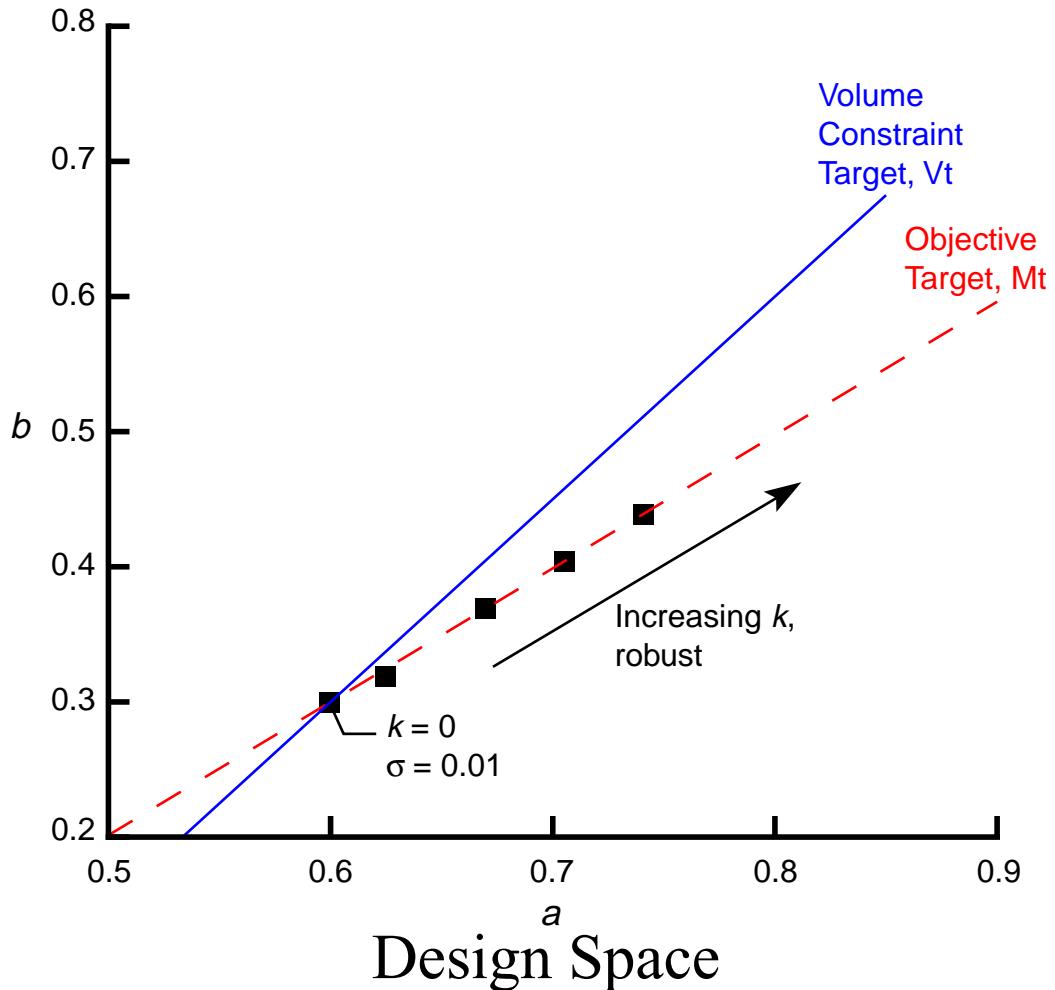


Design Space

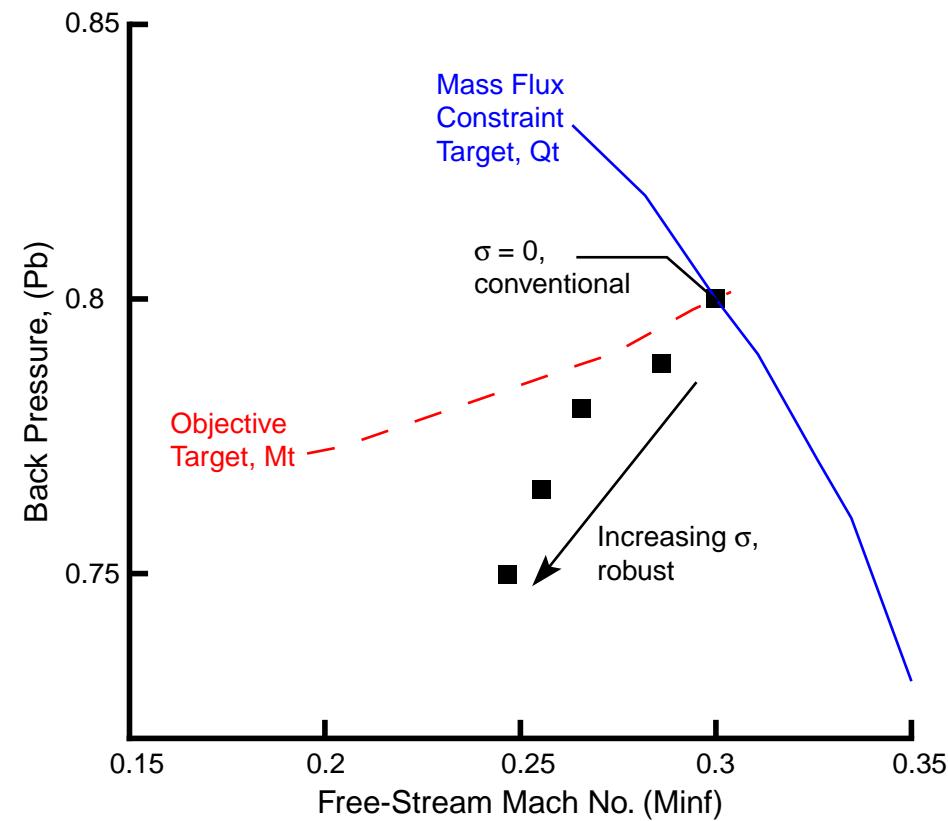


Nozzle Area Distributions

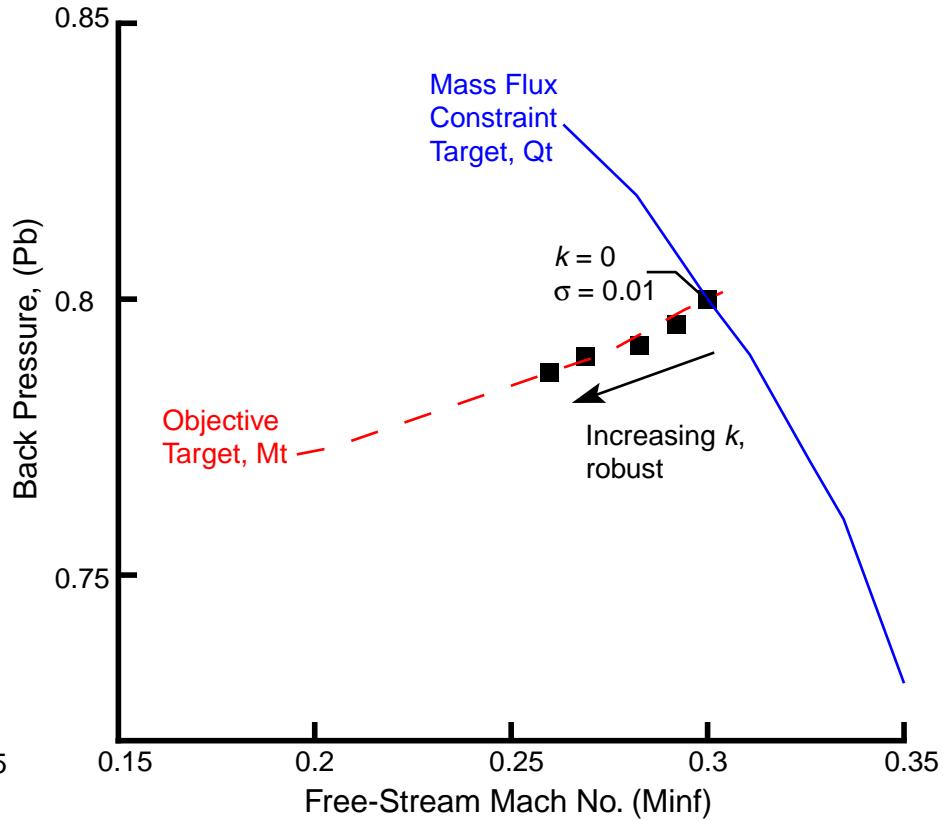
Robust Shape Optimization Results, Increasing k , $\sigma_a = \sigma_b = 0.01$



Robust Design for Flow Control Results



Results in Design Space
 $P_k = P_1 = 84.13 \%$



Results in Design Space
 $\sigma_{M_{\infty}} = \sigma_{P_b} = 0.01$

Quasi 1-D Study Conclusions

- The results demonstrate an implementation of the approximate statistical moment method for input uncertainty propagation and robust design in CFD.
- The approximate statistical moment method used appears feasible when considering robustness about input parameter mean values.
- The feasibility of this method using second-order sensitivity derivate must be assessed in two-or three-dimensional problems.